Article Adaptive Integral Sliding Mode based Path Following Control of Unmanned Surface Vehicle

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Abstract: This paper investigates the path following control problem for a unmanned surface vehicle (USV) in the presence of unknown disturbances and system uncertainties. The simulation study combines two different types of sliding mode surface based control approaches due to its precise tracking and robustness against disturbances and uncertainty. Firstly, an adaptive linear sliding mode surface algorithm is applied, to keep the yaw error within the desired boundaries and then an adaptive integral non-linear sliding mode surface is explored to keep an account of the sliding mode condition. Additionally, a method to reconfigure the input parameters in order to keep settling time, yaw rate restriction and desired precision within boundary conditions is presented. The main strengths of proposed approach is simplicity, robustness with respect to external disturbances and high adaptability to static and dynamics reference courses without the need of parameter reconfiguration.

Keywords: Unmanned Surface Vehicle; Guidance, Navigation and Control; Path Following; Adaptive Sliding Mode

1. Introduction

With the growing advancement in the sensor technology and navigation aids, USVs are becoming a popular tool in maritime domain for several applications ranging from environmental monitoring, military surveillance to scientific surveying and data collection. Mission oriented approach of USVs subject them to several types of maritime environment comprising of wind, wave and sea surface currents leading to requirement of designing and developing several autonomy levels for successful operation. Henceforth, design and development of approaches for Guidance, Navigation and Control (GNC) of a USV is an important research area for constructing operational and tactical approaches for seven different operational autonomy level of USVs as described by International Maritime Organisation (IMO).

Guidance and control of USV plays an important role in motion control system to manipulate the forces to enable a USV to follow a desired path whilst maintaining the stability. Three approaches, namely, **waypoint control**, **path following control** and **trajectory tracking** are generally considered in the domain of marine robotics to enable a USV to follow a designated path [1,2]:

- **Waypoint control:** In this strategy, Line of Sight (LOS) based approach is adopted to follow a certain waypoints, generated heuristically, in the required maritime environment.
- **Path following control:** In this strategy, a path generated through path planning algorithms is used as a reference, to be followed with no temporal constraints. Here, USV should converge and follow the desired path without any time constraints and simultaneously satisfies its assigned velocity profile.

() () • **Trajectory tracking:** In this strategy, temporal constraints are enforced upon the path generated using path planners. This is predominantly used with fully actuated marine vehicles reasoned with better manoeuvring capabilities.

1.1. State of the Art

The problem of **path following control** is highly non-linear in nature and has been studied from a perspective of observed disturbance control using sliding mode control (SMC) approach. The SMC problem for USVs, subjected to, higher order non linear operational disturbances, have been studied with varying control approaches like sliding mode [4–7]; fuzzy sliding mode [9]; proportional derivative fuzzy [10]; backstepping [11–14]; backstepping with adaptive radial basis function neural network [15]; sine function-based non-linear feedback [16]; hyperbolic tangent based nonlinear control [17]; sigmoid based nonlinear control [18]; function adaptive neural path following control [19]; model predictive control [20,21]; and non-linear feedback power functions [22].

In order to make control robust to disturbances and uncertainties, several approaches has been proposed in the SMC literature, see [23–33]. Some proposals of advanced sliding manifolds include recursive nonlinear sliding manifolds [34,35], non linear full order dynamics [36,37], sliding surfaces with adaptive damping parameters [38–40] and, in the last years, a vast collection of homogeneity based works, see [41] for instance. Applications of the properties of homogeneous systems is an important field of study in the current development of analysis and design of nonlinear controllers and observers. Homogeneity simplifies analysis and design of nonlinear control systems since the homogeneous vector fields have many properties similar to linear one and provides solutions with finite-time and fixed-time stability.

The dynamics generated by an homogeneous controller can be seen as a lineal dynamic system with an adaptive gain that grows to ∞ as $|x(t)| \rightarrow 0$, generating the well know singularity at the origin which is undesired for real applications. Nevertheless, as commented in [42], the practical implementation of homogeneous dynamics system designed in the continuous time domain prevents the use of explicit *Euler* discretization scheme to achieve a mere copy of the continuous time approach due to its simplicity. This type of discretization is considered inappropriate, especially when set-valued functions has to be considering, causing numerical chattering and sensitivity to the gains. As a result, without addressing the discretization issue, any comparison between homogeneous based solutions and other types of proposals may potentially lead to unfair conclusions

Based on the aforementioned results, in order to keep the discretization process simple, an adaptive lineal sliding mode surface law, that includes a nested integral sliding surface is introduced in this work. In this case, the dynamics flows with adaptive and finite damper gain, avoiding the effects of the peaking transient response inherent to linear systems and allowing fast responses at steady state, approximating the behaviour obtained with homogeneous solutions.

1.2. Major Contributions

The paper makes following contributions to the current state of existing approaches to SMC techniques for USVs:

- The proposed adaptive control approach is reconfigurable, without parametric adjustment, for various input trajectories and environmental disturbances of maritime environment, as it is shown through various simulation studies conducted in the manuscript.
- Because of the low-pass filtering properties related to the second order adaptive linear dynamics generated at the sliding variable, the cross over frequency (ω_c) of the system response can be used in the algorithm to estimate bound of the disturbance derivative. This implies that frequencies over ω_c does not affect the performance of the sliding variable response. From a practical implementation



Figure 1. 6 DOF motion representation with North-East-Down coordinate system (green) and body fixed reference frame (black).

perspective, this feature brings some advantages if the estimation of the maximum value of the disturbance derivative is a difficult task to solve.

- The proposed adaptive profile generates a low/high gain variation that depend on the absolute value of the error. This allows to avoid the saturation of the control input when the error is large (gain is small) and to create responses at steady state (gain is large) that generates fast disturbance compensation.
- The use of an scheme based on the adaptive placement of two poles related to
 a second order dynamical system with critical damped, allows to generate fast
 overdamped responses that avoids the introduction of considerable overshoots.

This paper has been structured as follows. First, in Section 2 we present the nonlinear dynamic model of the path following problem, the desired objectives to be achieved and a theoretical stability tool that is used in the posterior analysis of the control algorithm. Then, Section 3 describes the proposed adaptive integral sliding mode (AISM) algorithm. Results from numerical simulations are then presented and discussed in Section 4. Finally, conclusions are drawn in Section 5.

2. Problem Statement

The motion of the USV is shown in Figure 1, where a six degrees of freedom (DOF) model is presented. The earth fixed **Oo** is an inertial reference frame fixed to the earth's surface and the body fixed with origin **O** is a moving coordinate frame that it is fixed to the craft as in given in [1]. It is assumed an homogeneous mass distributed and xz-plane symmetrical, such that origin of the body fixed reference frame is chosen to be coincident with the center of the gravity. If we consider the path following problem the dynamics of heave, roll, and pitch can be neglected, so that the reduced model dynamics are given as

$$m(\dot{u} - vr - x_c r^2) = X \text{ (Surge)}$$
(1)

$$m(\dot{v} + ur + x_c \dot{r}) = Y$$
(Sway) (2)

$$I_z \dot{r} + m x_c (\dot{v} + ur) = N \text{ (Yaw)}$$
(3)

where m is the mass, u is the surge velocity, v the sway velocity, r the yaw rate, I_z the rotational inertia with respect to z axis, x_c is the x coordinate of the vehicle center in the fixed body reference frame and X,Y and N are the external forces and moments with respect to the surge, sway and yaw, respectively.

Assumption of constant forward speed and using the ship's *Norrbin* nonlinear mathematical model, see [43], implies that the steering equations of motion can be obtained as

$$\dot{\psi}(t) = r(t)$$

$$\dot{r}(t) = f(r) + g\delta(t) + d(t)$$
(4)

where, $\psi(t)$ is the yaw (orientation) angle, r(t) is the yaw rate, $\delta(t)$ is the rudder angle (the control variable to be designed) and d(t) is an unknown term to be compensated that includes parametric uncertainty and external disturbances (wind, waves, mobile loads). The dynamics functions are given as

$$g = -\frac{K}{T}$$

$$f(r) = -\frac{K}{T}H(r)$$

$$H(r) = a_1r + a_2r^3$$
(5)

where (*K*, *T*) are hydrodynamic coefficients and (a_1, a_2) are *Norrbin* coefficients.

In the path following problem it is required that the yaw angle ψ follows a reference angle ψ_r by means of the design of the rudder control signal $\delta(t)$. The following assumptions are taking account in this work.

Assumption 1. d(t) in (4) satisfies the following restriction

$$|d(t)| \leq d_{max}$$

with $\dot{d}_{max} > 0$ a positive real number.

Assumption 2. Henceforth, it is assumed that a reference yaw establish the desired input to be tracked, which can be obtained by means of path planning algorithms, that account for different environment constraints as in [44–47].

A dynamic reference model is used, in this work, to generate the desired course $(\psi_r(t), \psi_r(t), \ddot{\psi}_r(t))$.

The objective is to design a control law that creates overdamped responses with minimal overshooting (undershooting) and robustness properties for response of the yaw error, which is defined as

$$e(t) = \psi(t) - \psi_r(t) \tag{6}$$

In order to check the control performance of the proposed controller for the path following problem, we consider the following performance analysis indices mentioned in [12,14].

$$MAE = \frac{1}{t_{\infty} - t_0} \int_{t_0}^{\infty} |e(t)| dt$$
(7)

$$MIA = \frac{1}{t_{\infty} - t_0} \int_{t_0}^{\infty} |\delta(t)| dt$$
(8)

$$MTV = \frac{1}{t_{\infty} - t_0} \int_{t_0}^{\infty} |\delta(t) - \delta(t - \tau)| dt$$
(9)

where τ is the sampling time used in the simulation.

Furthermore, to check out the robustness properties of the solution, we compare the results with the algorithms proposed in [12,14] applying the following conditions:

 As in [12,14], we test two problems that uses two different types of reference input signals: step and sinusoidal.

- The tests includes results without disturbances (d(t) = 0) and with disturbances ($d(t) \neq 0$).
- The algorithm parameters are configured in the case of the step input reference without disturbances, such that all solutions provide the same value of the MIA index at the end of the test time.
- After that, the algorithms parameters are fixed and tested in the case of step with disturbances and in the case of the sinusoidal input reference. In this way we check the robustness of the solutions with respect to its capacity of adaptation to different scenarios from a specific parameter configuration.

The following theorem is introduced in order to analyse the stability properties of the AISM proposed solution.

Theorem 1. Consider the following cascade system

$$\dot{z}_1 = f_1(t, z_1) + g_1(t, z_1, z_2)z_2 \tag{10}$$

$$\dot{z}_2 = f_2(t, z_2)$$
 (11)

where $z_1 \in \mathbb{R}^n$, $z_2 \in \mathbb{R}^m$, $f_1(t, z_1)$ is continuously differentiable in (t, z_1) , and $f_2(t, z_1)$ and $g_1(t, z_1, z_2)$ are continuous and locally Lipschitz in z_2 and (z_1, z_2) , respectively.

The dynamics of (10) *when* $z_2 = 0$ *are*

$$\dot{z}_1 = f_1(t, z_1)$$
 (12)

If systems (12) and (11) are globally uniformly asymptotically stable (GUAS) and we know a C¹ Lyapunov function $V(t, z_1)$, two class- K_{∞} functions ϕ_1 and ϕ_2 , a class- $K \phi_3$ function and a positive semidefinite function $W(z_1)$ such that

$$\phi_1(||z_1||) \le V(t, z_1) \le \phi_2(||z_1||) \tag{13}$$

$$\frac{\partial V}{\partial t} + \frac{\partial V}{\partial z_1} f_1(t, z_1) \le -W(z_1) \tag{14}$$

$$||\frac{\partial V}{\partial z_1}|| \le \phi_3 \tag{15}$$

Besides, for each fixed z_2 there exists a continuous function $\zeta : \mathbb{R}^+ \to \mathbb{R}^+$ such that

$$\lim_{s \to \infty} \zeta(s) = 0 \tag{16}$$

$$|\frac{\partial V}{\partial z_1}g_1(t, z_1, z_2)|| \le \zeta(||z_1||)W(z_1)$$
(17)

Then we can conclude that the cascade system (10) and (11) is GUAS.

Proof. See [48] . □

3. Adaptive Integral Sliding Mode Surface Control Design

Derivation of e(t) in (6) leads to

$$\dot{e}(t) = r(t) - \dot{\psi}_r(t) \tag{18}$$

An adaptive sliding surface s(t) variable is defined as

$$s(t) = \dot{e}(t) + \lambda(e)e(t)$$
(19)

with $\lambda(e)$ a real positive time varying parameter.

Consider the integral term $\bar{s}(t)$

$$\bar{s}(t) = \int_0^t s(t)dt \tag{20}$$

Let's choose the control law as

$$\delta(t) = \frac{1}{g}(-f(r) + \ddot{\psi}_r(t) - \lambda(e)\dot{e}(t) - \dot{\lambda}(e)e(t) - \alpha(s,\bar{s})s(t) - \gamma(e)\bar{s}(t))$$
(21)

with $\lambda(e)$ defined as

$$\lambda(e) = max(\lambda_{min}, \lambda_{max} - (\frac{\lambda_{max} - \lambda_{min}}{|e(0)|})|e(t)|)$$
(22)

, the variable z(t), related a new sliding surface, defined as

$$z(t) = s(t) + \frac{\alpha}{2}\bar{s}(t)$$
(23)

and with the parameters $\alpha(s, \bar{s})$, $\gamma(\alpha)$ and $\delta(e)$ given as

$$\dot{\alpha}(s,\bar{s}) = \kappa |z|^{\delta} \operatorname{sign}(z) \operatorname{sign}(s)$$
(24)

$$\gamma(\alpha) = \frac{\alpha^2}{4} \tag{25}$$

$$\delta(e) = \left(\frac{\delta_{max} - \delta_{min}}{|e(0)|}\right)|e(t)| + \delta_{min}$$
(26)

Derivation of $\gamma(\alpha)$ and $\lambda(e)$ are given as

$$\dot{\gamma}(\alpha) = \frac{\alpha}{2}\dot{\alpha} \tag{27}$$

$$\dot{\lambda}(e) = \begin{cases} -\left(\frac{\lambda_{max} - \lambda_{min}}{|e(0)|}\right) sign(e(t))\dot{e}(t) & \text{if } \lambda > \lambda_{min} \\ 0 & \text{if } \lambda \le \lambda_{min} \end{cases}$$
(28)

The control algorithm is designed by an appropriate selection of the parameters λ_{max} , λ_{min} , $\alpha(0)$, κ , δ_{max} and δ_{min} , as it will be introduced in the numerical simulations section.

Theorem 2. Consider the ship course dynamics described in (4) that complies with assumption 1. The application of the control law (21) to dynamic system (4) implies that the closed compact set Ω_e defined as

$$\Omega_e = \{ (e(t), \dot{e}(t)) \in \mathbb{R}^2 : |e(t)| < \frac{\mu}{|\cos(\theta)||\sin(\vartheta)|} \land |\dot{e}(t)| < \frac{\mu}{|\cos(\theta)||\cos(\vartheta)|} \}$$
(29)

is GUAS with μ , θ and ϑ given as

$$\mu = \sqrt[\delta+1]{\frac{\dot{d}_{max}}{\kappa}} \tag{30}$$

$$\theta = \operatorname{atan}(\lambda) \tag{31}$$

$$\vartheta = \operatorname{atan}(\frac{\lambda}{2}) \tag{32}$$

Proof. Application of control law (21) to dynamic system (4) creates the following cascade system.

$$\dot{e}(t) = -\lambda e(t) + s(t) \tag{33}$$

$$\dot{s}(t) = -\alpha s(t) - \gamma \bar{s}(t) + d(t) \tag{34}$$

The dynamics of $\dot{e}(t)$ when s(t) = 0 (dynamics of the yaw error at the sliding condition) are

$$\dot{e}(t) = -\lambda e(t) \tag{35}$$

with $\lambda > 0$. Therefore system (35) is GUAS, with exponential convergence. Derivation of $\dot{s}(t)$ leads to

$$\ddot{s}(t) + \alpha \dot{s}(t) + \gamma s(t) + \dot{\alpha} s(t) + \dot{\gamma} \bar{s}(t) + \dot{d}(t) = 0$$

From (25), (27) and (23) it is obtained

$$\ddot{s}(t) + \alpha \dot{s}(t) + \frac{\alpha^2}{4}s(t) + \dot{\alpha}z(t) + \dot{d}(t) = 0$$

Substitution of $\dot{\alpha}$ from (24) implies that the second order dynamics equation related to s(t) is

$$\ddot{s}(t) + \alpha \dot{s}(t) + \frac{\alpha^2}{4} s(t) + \kappa |z|^{\delta+1} \operatorname{sign}(s) + \dot{d}(t) = 0$$
(36)

Applying assumption 1, $|z| > \sqrt[\delta+1]{\frac{d_{max}}{\kappa}}$ implies that

$$\kappa |z|^{\delta+1} \operatorname{sign}(s) + \dot{d} = \rho_z s \tag{37}$$

with $\rho_z > 0$. Therefore the characteristic polynomial of (36) is *Hurwitz* for all $z(t) \notin \Omega_z$ where

$$\Omega_z = \{ z(t) \in \mathbb{R} : |z(t)| < \mu \}$$
(38)

with μ defined in (30).

This implies that (34) is GUAS with respect to the closed set Ω_z . Note that dynamics in (36) can be viewed as a second order linear dynamics with adaptive critical damping (exponential convergence related to the fastest response with no overshooting), being perturbed by the overestimation $\rho_z s$ caused by the compensation of the unknown term.

Inside Ω_z we have that

$$|s(t) + \frac{\alpha}{2}\bar{s}(t)| < \mu$$

which geometrically entails:

$$|\bar{s}(t)| < \frac{\mu}{|\sin(\vartheta)|} \tag{39}$$

$$|s(t)| < \frac{\mu}{|\cos(\vartheta)|} \tag{40}$$

with ϑ defined in (32).

Inside Ω_s we have that

$$|\dot{e}(t) + \lambda e(t)| < \frac{\mu}{|\cos(\vartheta)|}$$

Following the previous approach implies that:

$$|e(t)| < \frac{\mu}{|\cos(\vartheta)||\sin(\theta)|}$$
(41)

$$|\dot{e}(t)| < \frac{\mu}{|\cos(\vartheta)||\cos(\theta)|} \tag{42}$$

with θ defined in (31).

Applying Theorem 1 with

$$\begin{split} \phi_1(||e||) &= k_1 e^2 \\ \phi_2(||e||) &= k_2 e^2 \\ \phi_3(||e||) &= k_3 |e| \\ W(e) &= k_4 e^2 \\ V(e) &= \frac{1}{2} e^2 \\ \zeta(e) &= \frac{k_5}{k_4 |e|} \end{split}$$

where $k_1 < 0.5$, $k_2 > 0.5$, $k_3 > 1.0$, $k_4 < \lambda$ and $k_5 > 1.0$, entails that cascade system given in (33) and (34) is GUAS with respect to the closed compact sets Ω_e and Ω_s , respectively. \Box

Let's note that

The size of Ω_e , Ω_s and Ω_z depend on adaptive values of λ , α and δ .

Ф

 $|z(t)| \rightarrow 0$ implies that $\dot{\alpha} \rightarrow 0$, so the value of α is stabilised at the steady-state of z(t), that is, inside Ω_z .

4. Numerical simulations

In this section we introduce numerical simulations of the path following problem with parameters given in Table 1 and being executed under the following assumption.

Assumption 3. The numerical simulations are executed using the explicit Euler method with fixed sampling time $\tau = 0.1 s$.

Parameter	Value
K	0.21
Т	107.76
<i>a</i> ₁	13.17
<i>a</i> ₂	16323.46

Table 1: Model parameters.

4.1. Constant yaw reference

This test is presented in [14] with a required a change in the yaw orientation angle from zero initial condition up to 50 degrees assuming that d(t) = 0. Table 2 show the parameters used in [14]. Based on this results the parameter a_2 of the synergetic controller

Parameter	Value
k_1	0.0017
ω	0.6000

Table 2: Nonlinear concise backstepping controller parameters.

9 of 17

presented in [12] is changed to achieve the same MIA at the end of the simulation. Table 3 show the parameters used with this algorithm.

Parameter	Value
<i>a</i> ₁	0.090
<i>a</i> ₂	1.891
T_1	28.000

Table 3: Synergetic controller parameters.

The parameters of the AISM algorithm are obtained as follows

- Consider a settling time $t_s = 150s$, a maximum desired yaw rate $r_{max} = \frac{0.70\pi}{180}$ degrees per second and a required precision $\epsilon = 1.0e 3$.
- The value of *α*(0) is obtained assuming an exponential convergence of the error from initial condition *e*(0) to desired precision *ε* with a desired settling time *t_s*

$$\alpha(0) = \frac{-log(\frac{\epsilon}{|e(0)|})}{t_s} = 0.0451$$
(43)

• The value of λ_{min} is related to the initial conditions of the problem and the maximum desired yaw rate as

$$\lambda_{min} = \frac{r_{max}}{|e(0)|} = 0.014 \tag{44}$$

and λ_{max} is calculated as

$$\lambda_{max} = 2.0\lambda_{min} = 0.028\tag{45}$$

• The value of κ must be higher than \dot{d}_{max} in order to obtain a small value for μ . Because of the low-pass filtering properties of (36), the value of \dot{d}_{max} can be further refined by estimating the cross over frequency $\omega_c(t)$ of the second order system related to s(t)

$$\omega_c(t) = \frac{2}{\alpha(t)} \tag{46}$$

Therefore κ is calculated as an adaptive gain that takes account of ω_c and the desired precision

$$\kappa = \frac{\omega_c(t)}{\epsilon} \tag{47}$$

• The values of δ_{min} and δ_{max} are related by means of the condition $\delta_{max} = 2.0 \delta_{min}$. The value of δ_{min} is adjusted with simulations such that the value of the performance index MIA is equal, at the end of test time, to the value obtained with benchmark selected controllers. This choice leads to the following numerical values of δ_{min} and δ_{max}

$$\delta_{min} = 1.76952 \tag{48}$$

$$\delta_{max} = 3.53904 \tag{49}$$

This condition generates an adequate adaption of the value of δ that allows to obtain the desired low/high gain profile with respect to the absolute value of e(t).

States and control effort are provided in Figure 2 where it is observed that all the solutions provide a similar setting time. Although the evolution of the yaw error is



Figure 2. Constant yaw reference test with d(t) = 0. States and control. Cyan line : Reference ; Red line: Concise backstepping (Zhang et al.); Blue line: Synergetic (Muhammad et al.); Black line: Adaptive sliding mode (González-Prieto et al.)

similar, it can be observer in the detail of the control effort that the evolution of the rudder angle is quite different compared with previous algorithms.

Time evolution of performance indices are shown in Figure 3, with a detailed view of MIA performance index at the end of the test, and final numerical values in Table 4. Figure 4 shows the evolution of the adaptive parameters used at the proposed AISM algorithm.

Algorithm	MAE	MIA	MTV
Concise Backstepping [14]	0.042227	0.011348	3.6396e-5
Synergetic [12]	0.035641	0.011348	6.4436e-5
AISM	0.038557	0.011348	4.6133e-5

Table 4: Constant yaw reference test with d(t) = 0. Performance indices.

Next, in order to test the robustness of the algorithms, the following disturbance is considered in (4)

$$d(t) = D[\cos(\omega_d t) + 0.83 \sin(3.29\omega_d t - 0.14) + 1.23 \cos(8.12\omega_d t + 0.26) + 0.65 \sin(1.37\omega_d t + 0.36)e^{\cos(2.21\omega_d t + 0.13)}]$$
(50)

with

$$D = 0.0025$$
 (51)

$$\omega_d = 0.0703 \frac{rad}{s} \tag{52}$$



Figure 3. Constant yaw reference test with d(t) = 0. Performance indices evolution.Cyan line : Reference ; Red line: Concise backstepping (Zhang et al.); Blue line: Synergetic (Muhammad et al.); Black line: Adaptive sliding mode (González-Prieto et al.)



Figure 4. Constant yaw reference test with d(t) = 0. Adaptive parameters evolution.



Figure 5. Constant yaw reference test with $d(t) \neq 0$. States and control.Cyan line : Reference ; Red line: Concise backstepping (Zhang et al.); Blue line: Synergetic (Muhammad et al.); Black line: Adaptive sliding mode (González-Prieto et al.)

States and control effort are provided in Figure 5 where it is clear that the proposed AISM cancels the effect of the external disturbance, keeping the desired performance at steady-state generating a rudder angle control that provides fast response attenuation without generation of overshooting. Figure 6 shows the evolution of the sliding variable s(t) and the external disturbance d(t) introduced to test robustness properties of the compared algorithms.

4.2. Sinusoidal yaw reference

In this case, as in [14], the yaw reference to follow is a sinusoidal signal defined as

$$\omega_d = \frac{50\pi}{180.0} \sin(\frac{2\pi}{600.0}) \tag{53}$$

where the initial yaw angle is

$$\psi(0) = \frac{10\pi}{180.0} \tag{54}$$

States and control effort are provided in Figure 7 where it is clear that AISM is capable to follow the yaw reference with no appreciable delay keeping the desired settling time. As in the previous test, results with sinusoidal reference are tested introducing disturbance (50). Figure 8 shows the states and control effort obtained in this case, where, as in the constant reference test, the steady-state performance and the settling time obtained with AISM are preserved despite the presence of the external unknown disturbance.

5. Conclusions and future works

In this work we have proposed an approach to develop an adaptive integral sliding mode procedure to design a nonlinear controller for the path following of surface



Figure 6. Constant yaw reference test with $d(t) \neq 0$. Sliding mode variable s(t) and external disturbance d(t).



Figure 7. Sinusoidal yaw reference test with d(t) = 0. States and control.Cyan line : Reference ; Red line: Concise backstepping (Zhang et al.); Blue line: Synergetic (Muhammad et al.); Black line: Adaptive sliding mode (González-Prieto et al.)



Figure 8. Sinusoidal yaw reference test with $d(t) \neq 0$. States and control.

vehicles. The solution has been proposed based on the application of adaptive gains that changes the dumping properties of the sliding surfaces and derives in a low/high gain profile such that it allows to overcome the use of large control inputs at initial conditions, keeping a desired higher gain at steady state.

The results obtained in the numerical simulations shows that the proposed AISM algorithm achieves the desired performance with fixed and time varying references cancelling the effect of the external disturbances. The performance is evaluated with a fixed parameter configuration that can be obtained from a settling time, maximum allowable yaw rate and steady state precision. The algorithm achieves the desire response without the need of develop a new parameter configuration for each type of test, showing its robustness properties.

An advantage of the method is its robustness with respect to an overestimation of \dot{d}_{max} : the performance is not highly degraded if this bound is not accurately known. However, choosing an overly large value might cause oscillations in the response of the estimation error.

Another important aspect about the presented solution is its practical approach, such that a desired precision value of the yaw error at steady state is integrated in the design procedure in order to configure control parameters.

A deepest research of the functions that can be used to define the adaptive values of $\lambda(e)$ and $\delta(e)$ is an interesting open problem that can be analysed from the perspective of model predictive control in order to integrate an optimal point of view in the design of the adaptive parameters.

The extension of this procedure with the assumption of partial state feedback will be addressed in future researches by means of the application of an adaptive integral sliding mode observers.

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Abbreviations

The following abbreviations are used in this manuscript:

- USV Unmanned Surface Vehicle
- IMO International Maritime Organisation
- LOS Line of Sight
- SMC Sliding Mode Control
- AISM Adaptive Integral Sliding Mode
- GUAS globally uniformly asymptotically stable
- MAE Mean Absolute Error
- MIA Mean Integral Absolute
- MTV Mean Total Variation

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